

Service Bulletin

Fuselage - Primary Structure, Center-Mid Fuselage - Routing Modification of the VFG Power-Feeder Harnesses

BD500-534101
Issue No. 008

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Manufacturer:

AIRBUS

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The listed documents are included in Issue 008, dated 2024-03-13, of this publication.

Document title	Data module code	Issue date	Applicable to
Fuselage - Primary Structure, Center-Mid Fuselage - Routing Modification of the VFG Power-Feeder Harnesses - Service Bulletin	BD500-A-J53-41-01-00AAA-930A-A	2024-03-13	50010-50047, 55003-55070
Routing Modification of the VFG Power-Feeder Harnesses - Material information	BD500-A-J53-41-01-01AAA-934A-A	2024-03-13	50010-50047, 55003-55070
PART A - Inspection of the VFG Power-Feeder Harnesses - Accomplishment instruction	BD500-A-J53-41-01-01AAA-933A-A	2024-03-13	50010-50017, 55003-55023, 55025
PART B - Installation of Clamps and Routing Modification of the VFG Power-Feeder Harnesses - Accomplishment instruction	BD500-A-J53-41-01-02AAA-933A-A	2024-03-13	50010-50017, 55003-55023, 55025
PART C - Installation of Brackets and Supports and Routing Modification of the VFG Power-Feeder Harnesses - Accomplishment instruction	BD500-A-J53-41-01-03AAA-933A-A	2024-03-13	50010-50047, 55003-55070

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Fuselage - Primary Structure, Center-Mid Fuselage - Routing Modification of the VFG Power-Feeder Harnesses - Service Bulletin

Applicability: 50010-50047, 55003-55070

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Data Module/Technical Publication	Title
BD500-A-J53-41-01-01AAA-934A-A	Routing Modification of the VFG Power-Feeder Harnesses - Material information
BD500-A-J53-41-01-01AAA-933A-A	PART A - Inspection of the VFG Power-Feeder Harnesses - Accomplishment instruction
BD500-A-J53-41-01-02AAA-933A-A	PART B - Installation of Clamps and Routing Modification of the VFG Power-Feeder Harnesses - Accomplishment instruction
BD500-A-J53-41-01-03AAA-933A-A	PART C - Installation of Brackets and Supports and Routing Modification of the VFG Power-Feeder Harnesses - Accomplishment instruction

Service Bulletin

Common information

THIS SERVICE BULLETIN IS SENT TO EACH OPERATOR. IF THE OPERATOR HAS LEASED AIRCRAFT, SEND THE SERVICE BULLETIN TO THE LESSEE. IF THE OPERATOR HAS SOLD AIRCRAFT OR TRANSFERRED SPARES IN THE LAST SIX MONTHS, SEND THIS SERVICE BULLETIN TO THE NEW OWNER, UNLESS CONFIRMATION WAS RECEIVED THAT AIRBUS HAS INCLUDED THE NEW OPERATOR ON THE DISTRIBUTION LIST.

Management information

Compliance category: Mandatory (Service bulletin must be accomplished)

Task category: Inspection and modification

Original issue date: 2018-04-19

Impact list

Impact (Weight and balance):

Refer to the Weight and balance section of the Planning information section.

Impact (Maintenance publications):

Refer to the Documentation affected section of the Planning Information section.

Accomplishment limits

Time compliance: Basic limit

Remarks:

Refer to the Compliance section of the Planning Information section.

Replacement service bulletin

None

Revision information

Revision information

The revision information letter transmits:

Issue 008 of Service Bulletin BD500-534101, dated 2024-03-13.

Additional work

The changes in this revision have no effect on aircraft that have incorporated a previous issue of this service bulletin.

Revision reason

Revised the Management information to:

Update the Compliance category to Mandatory.

Revised the Planning information to:

Change the Compliance section to Mandatory.

Update the Referenced Documentation section.

Update the Industry support information section.

Revised the Accomplishment instructions to:

Add general information in PART C.

Update some steps in PART C.

This revision can also include small editorial changes that do not have an effect on the technical content.

Revision history

Issue 001, dated 2018-04-19
Issue 002, dated 2018-05-11
Issue 003, dated 2018-08-14
Issue 004, dated 2018-10-02
Issue 005, dated 2020-06-12
Issue 006, dated 2020-08-14
Issue 007, dated 2020-10-02

Revision sequence

Discard the copy of Service Bulletin BD500-534101, Issue 007, dated 2020-10-02. Replace with the copy attached.

Summary

Applicability

PART A and PART B

BD-500-1A10, Aircraft 50010 thru 50017

BD-500-1A11, Aircraft 55003 thru 55023 and 55025

PART C

BD-500-1A10, Aircraft 50010 thru 50047

BD-500-1A11, Aircraft 55003 thru 55070

Reason

It is possible that the power-feeder harnesses of the Variable Frequency Generator (VFG) rub against the wheel bins. This can cause damage to the harnesses.

Description

This service bulletin is divided into three parts: PART A, PART B and PART C.

PART A - Inspection of the VFG Power-Feeder Harnesses

The wheel bins are removed. A visual inspection is done on the VFG power-feeder harnesses and the wheel bins. If no damage is found, do PART B or PART C. If damage is found, call the CRC for disposition and do PART B or PART C.

PART B - Installation of Clamps and Routing Modification of the VFG Power-Feeder Harnesses

If necessary, the wheel bins are removed. Clamps are added to the VFG power-feeder harnesses to prevent rubbing. The wheel bins are installed.

PART C - Installation of Brackets and Supports and Routing Modification of the VFG Power-Feeder Harnesses

The wheel bins are removed. Brackets and supports are added to the VFG power-feeder harnesses to prevent rubbing. The wheel bins are installed.

Concurrent requirements

None

Manpower

It will take an estimated 22 h to complete PART A of this service bulletin.

It will take an estimated 3 h to complete PART B of this service bulletin.

It will take an estimated 26 h to complete PART C of this service bulletin.

Material information

BD500-A-J53-41-01-01AAA-934A-A

Planning information

Applicability

Aircraft affected:

PART A and PART B

AIRCRAFT MODEL	SERVICE AIRCRAFT SERIAL NUMBER	PRODUCTION AIRCRAFT SERIAL NUMBERS
BD-500-1A10	50010 thru 50017	50007 thru 50009, and 50018 thru 50047 ¹
BD-500-1A11	55003 thru 55023, 55025	55024 and 55026 thru 55070 ¹
1 Aircraft will have an equivalent modification incorporated before delivery.		

PART C

AIRCRAFT MODEL	SERVICE AIRCRAFT SERIAL NUMBER	PRODUCTION AIRCRAFT SERIAL NUMBERS
BD-500-1A10	50010 thru 50047	50048 and subsequent serial number(s) ¹
BD-500-1A11	55003 thru 55070	55071 and subsequent serial number(s) ¹
1 Aircraft will have an equivalent modification incorporated before delivery.		

Note

The instructions in this service bulletin are only applicable to the systems and parts installed at the time of delivery of the aircraft or as changed by Airbus Service Bulletins. Before you do this service bulletin, examine all Supplemental Type Certificates (STCs), Supplemental Type Approvals (STAs), or equivalent action changes to make sure that you can complete this service bulletin.

Reason

Problem and effect

It is possible that the power-feeder harnesses of the VFG rub against the wheel bins. This can cause damage to the harnesses.

Objective/Solution

This service bulletin gives the procedure to do a visual inspection of the VFG power feeder-harnesses and wheel bins, and to add clamps or brackets and supports for the VFG power feeder-harnesses to prevent chaffing.

Description

This service bulletin is divided into three parts: PART A, PART B and PART C.

PART A - Inspection of the VFG Power-Feeder Harnesses

The wheel bins are removed. A visual inspection is done on the VFG power-feeder harnesses and the wheel bins. If no damage is found, do PART B or PART C. If damage is found, call the CRC for disposition and do PART B or PART C.

PART B - Installation of Clamps and Routing Modification of the VFG Power-Feeder Harnesses

If necessary, the wheel bins are removed. Clamps are added to the VFG power-feeder harnesses to prevent rubbing. The wheel bins are installed.

PART C - Installation of Brackets and Supports and Routing Modification of the VFG Power-Feeder Harnesses

The wheel bins are removed. Brackets and supports are added to the VFG power-feeder harnesses to prevent rubbing. The wheel bins are installed.

Compliance

Mandatory Service Bulletin

Applicability: 50010-50017, 55003-55023, 55025

For PART A and PART B

Airbus recommends that PART A and PART B of this service bulletin be done at no more than 935 flight hours, or no later than five months for pre REO 500-20-10-001 or 1870 flight hours, or no later than ten months for post REO 500-20-10-001, from the original release date on this service bulletin, whichever comes first, unless otherwise directed by the airworthiness authority of the operator.

Applicability: 50010-50047, 55003-55070

For PART C

Airbus recommends that PART C of this service bulletin be done before the aircraft has 9350 flight hours, unless otherwise directed by the airworthiness authority of the operator

Note

If it is not possible to complete all the instructions in this service bulletin because of the configuration of the aircraft, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.

Note

This service bulletin does not affect Airworthiness Limitations Items (ALIs) or Damage Tolerance Inspections (DTIs).

Concurrent requirements

None

Approval

The technical content of this service bulletin has been approved under the authority of the TC-CA Design Approval Organization No: DAO #19-Q-02.

Manpower

The man-hours given below are estimates for direct labor done by an experienced crew on an aircraft prepared for maintenance.

These estimates do not include the time necessary for:

- Administrative functions such as planning, liaising, familiarization, and report writing.
- Non-productive elapsed time such as rest time, having meals, and crew shift changes.
- Elapsed time such as for paint to dry, sealant to cure, and fuel tank venting.
- Preparation for modification (unless specified in the Accomplishment Instructions) such as cleaning and getting parts, tools, consumables, and ground equipment.
- Quality assurance inspections, troubleshooting and/or correction of discrepancies found when a task is done.
- All other activities that are not considered by Airbus as directly related to the accomplishment of this procedure.

Applicability: 50010-50017, 55003-55023, 55025

It will take an estimated 22 h to complete PART A of this service bulletin. The man-hours are divided as shown in the table that follows:

TASK	MAN-HOURS
Job Set-Up	10.00 h
Procedure	2.00 h
Test	0 h
Job Close-Up	10.00 h

It will take an estimated 3 h to complete PART B of this service bulletin. The man-hours are divided as shown in the table that follows:

TASK	MAN-HOURS
Job Set-Up	0.25 h
Procedure	2.50 h
Test	0 h
Job Close-Up	0.25 h

Applicability: 50010-50047, 55003-55070

It will take an estimated 26 h to complete PART C of this service bulletin. The man-hours are divided as shown in the table that follows:

TASK	MAN-HOURS
Job Set-Up	10.00 h
Procedure	6.00 h
Test	0 h
Job Close-Up	10.00 h

Weight and balance

PART A and PART B

None

For PART C

BD-500-1A10

WEIGHT	ARM	MOMENT ¹
+ 1.04 lb	899.40 in.	+0.9354 (1000 lb•in)
+ 0.47 kg	22.84 m	+0.0107 (1000 kg•m)
1 The value in the Moment column is divided by 1000. Refer to Weight and Balance Manual.		

BD-500-1A11

WEIGHT	ARM	MOMENT
+ 1.04 lb	899.40 in.	+0.9354 (1000 lb•in)
+ 0.47 kg	22.84 m	+0.0107 (1000 kg•m)

Electrical load data

Not changed

Referenced documentation

Refer to the SB Introduction document BD500-000000 for definition or explanation of service bulletin content.

Modification Summary 500T102025, Revision A-1

Modification Summary 500T102157, Revision D

For Airbus reference only: Restriction and/or Special Instruction (RSI) C-500-000-91-0002, Revision B

For Airbus reference only: Restriction and/or Special Instruction (RSI) C-500-000-91-0004, Revision A

For Airbus reference only: PCR 53648

Documentation affected

1 Maintenance Publications

Aircraft Maintenance Publication (AMP)

Illustrated Part Data Publication (IPDP)

2 Operational Publications

None

Industry support information

This service bulletin is part of an AIRBUS monitored retrofit campaign. For detailed information, please refer to Retrofit Information Letter (RIL) S91M23002626. To organize this retrofit, please contact AIRBUS retrofit management department at: a220monitored.retrofit@airbus.com.

Material information

BD500-A-J53-41-01-01AAA-934A-A

Accomplishment instructions

BD500-A-J53-41-01-01AAA-933A-A

BD500-A-J53-41-01-02AAA-933A-A

BD500-A-J53-41-01-03AAA-933A-A

Additional information

None

Routing Modification of the VFG Power-Feeder Harnesses - Material information

Applicability: 50010-50047, 55003-55070

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Table 1 References

Data Module/Technical Publication	Title
BD500-A-J51-31-00-01AAA-070A-A	Supply requirements - Consumables and materials

Service Bulletin

Material set list

No information

Support equipment list

For a list of supplies, refer to the different tasks that are given as references in this service bulletin.

Supplies list

None

Spares list

The Spare Sets in the table that follows are available from Airbus Canada:

For PART B

Applicability: 50010-50017

Spare set: Kit BD500534101K1

Name: KIT, VFG POWER FEEDER CLAMPS

Part number [Mfg. code]	Name	Quantity	Remarks
C01384564-N0005 [3AB48]	Packer, Harness Saddle Clamps	2	



Part number [Mfg. code]	Name	Quantity	Remarks
AS21919WCH22 [81343]	Clamp Loop, Cushioned	2	
AS21919WCH20 [81343]	Clamp Loop, Cushioned	2	
TA09203CO22HC [84971]	Clamp, Saddle	2	
B0201081CK3-9 [3AB48]	Screw, Hex HD	3	
B0202033C0316K [3AB48]	Washer, Flat	9	
NAS1291C3M [80205]	Nut, Self-Lkg, Hex	3	
B0201081CK3-16 [3AB48]	Screw, Hex HD	4	
B0201081CK3-17 [3AB48]	Screw, Hex HD	2	

Applicability: 55003-55023, 55025

Spare set: Kit BD500534101K2
Name: KIT, VFG POWER FEEDER CLAMPS

Part number [Mfg. code]	Name	Quantity	Remarks
C01384564-N0006 [3AB48]	Packer, Harness Saddle Clamps	2	
AS21919WCH22 [81343]	Clamp. Loop, Cushioned	2	
AS21919WCH20 [81343]	Clamp. Loop, Cushioned	2	
TA09203CO22HC [84971]	Clamp, Saddle	2	
B0201081CK3-9 [3AB48]	Screw, hex HD	3	
B0202033C0316K [3AB48]	Washer, Flat	9	
NAS1291C3M [80205]	Nut, Self-Lkg, Hex	3	

Part number [Mfg. code]	Name	Quantity	Remarks
B0201081CK3-16 [3AB48]	Screw, Hex HD	4	
B0201081CK3-17 [3AB48]	Screw, Hex HD	2	

For PART C

Applicability: 50010-50047, 55003-55070

Spare set: Kit BD500534101K3

Name: KIT, VFG POWER FEEDER BRACKETS

Part number [Mfg. code]	Name	Quantity	Remarks
C01368480-001 [K4585]	Bracket Assy, Elec Sys, WTBF	1	
C01368417-001 [K4585]	Support Assy, Elec Sys, WTBF	1	
C01368481-001 [K4585]	Bracket Assy, Elec Sys, WTBF	1	
C01368479-001 [K4585]	Bracket Assy, Elec Sys, WTBF	2	
C01368421-001 [K4585]	Support Assy, Elec Sys, WTBF	1	
C01368422-001 [K4585]	Support Assy, Elec Sys, WTBF	1	
C01368423-001 [K4585]	Support Assy, Elec Sys, WTBF	1	
B0202033C0316K [3AB48]	Washer, Flat	2	
MS20470AD4-5A [80205]	Rivet, Solid, Univ HD	14	
B0201081CK3-8 [3AB48]	Screw, Hex HD	4	
TA09203CO20HC [84971]	Clamp, Saddle	2	
TA09201CO08HC [84971]	Clamp, Saddle	1	

Part number [Mfg. code]	Name	Quantity	Remarks
MS20470AD4-4A [80205]	Rivet, Solid, Univ HD	6	
B0201081CK3-9 [3AB48]	Screw, Hex HD	4	
B0201081CK3-10 [3AB48]	Screw, Hex HD	2	

Procurement information:

Procurement address:

Name: For customers based in the Americas

Business unit:

Business unit name: Satair USA, Inc

525 Westpark Drive, Suite 400

Peachtree City, GA 30269

United States of America

Ph.: 1 404 675 6333

Please contact your Customer Order Specialist

<https://www.satair.com/market/search>

CPHSB7X

Name: For customers based in Europe, Middle East, Africa

Business unit:

Business unit name: Satair A/S

Amager Landevej 147A

2770 Kastrup

Denmark

Ph.: 45 3247 0100

Please contact your Customer Order Specialist

<https://www.satair.com/market/search>

CPHSB7X

Name: For customers based in the Asia Pacific (excluding mainland China)

Business unit:

Business unit name: Satair Pte Ltd

12 Seletar Aerospace Link

Singapore 797553

The Republic of Singapore

Ph.: 65 6543 0977

Please contact your Customer Order Specialist

<https://www.satair.com/market/search>

CPHSB7X

Name: For customers based in mainland China

Business unit:

Business unit name: Satair (Beijing) Co. Ltd

Tian Zhu Lu 8, Tianzhu Airport Industrial Zone

Beijing 101312

People's Republic of China

Ph.: 86 10 8048 6161

Please contact your Customer Order Specialist

<https://www.satair.com/market/search>
CPHSB7X

Removed spares list

None

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PART A - Inspection of the VFG Power-Feeder Harnesses - Accomplishment instruction

Applicability: 50010-50017, 55003-55023, 55025

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References

Table 1 References

Data Module/Technical Publication	Title
BD500-A-J12-00-00-01AAA-398A-A	Standard configuration for maintenance - Aircraft system configuration
BD500-A-J53-82-84-07AAA-520A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J53-82-84-08AAA-520A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J53-82-84-07AAA-720A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Install procedure
BD500-A-J53-82-84-08AAA-720A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Install procedure

Preliminary requirements

Required conditions

Table 2 Required conditions

Action/Condition	Data Module/Technical publication
None	

Support equipment

Table 3 Support equipment

Name	Identification/Reference	Quantity	Remark
None			

Consumables, materials, and expendables

Table 4 Consumables, materials, and expendables

Name	Identification/Reference	Quantity	Remark
None			

Spares

Table 5 Spares

Name	Identification/Reference	Quantity	Remark
None			

Safety conditions

None

Procedure

- 1 Job set-up

Note

The steps in the Job set-up section of this service bulletin are recommended steps. The steps give a recommendation to get access to the work area. This

recommendation is to give a safe work area and to minimize possible damage to surrounding aircraft parts. Alternative steps can be used at the discretion of the operator.

- 1.1 Make sure that the aircraft is in standard configuration for maintenance (refer to AMP BD500-A-J12-00-00-01AAA-398A-A).

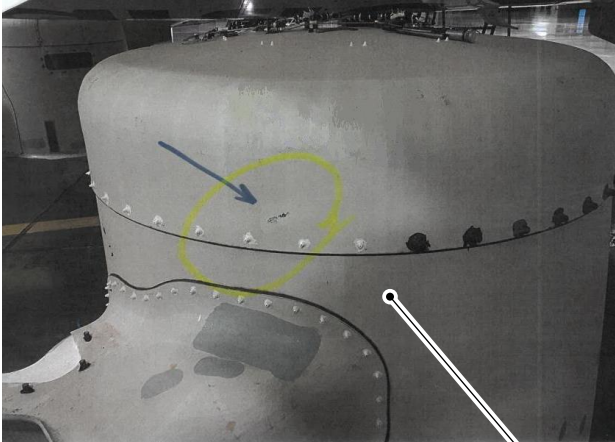
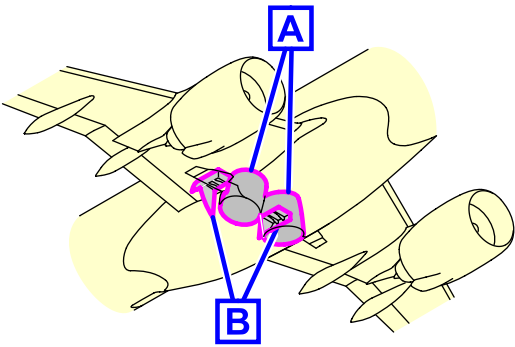
2 Procedure

- 2.1 If it is not possible to complete all the instructions in this service bulletin because of the configuration of the aircraft, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.
- 2.2 Remove the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-520A-A).
- 2.3 Remove the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-520A-A).
- 2.4 Do a visual inspection for damage at the intersection of the Variable Frequency Generator (VFG) power-feeder harnesses and the surface of the wheel bins, left and right sides as follows:

Refer to Fig. 1

- 2.4.1 If you find damage on the VFG power-feeder harnesses and/or the wheel bins on the left and/or right side, speak with a Customer Response Center (CRC) representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus.
- 2.4.2 If no damage is found on the VFG power-feeder harnesses and/or the wheel bins, do PART B or PART C of the service bulletin.
- 2.4.3 If necessary, install the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-720A-A).
- 2.4.4 If necessary, install the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-720A-A).

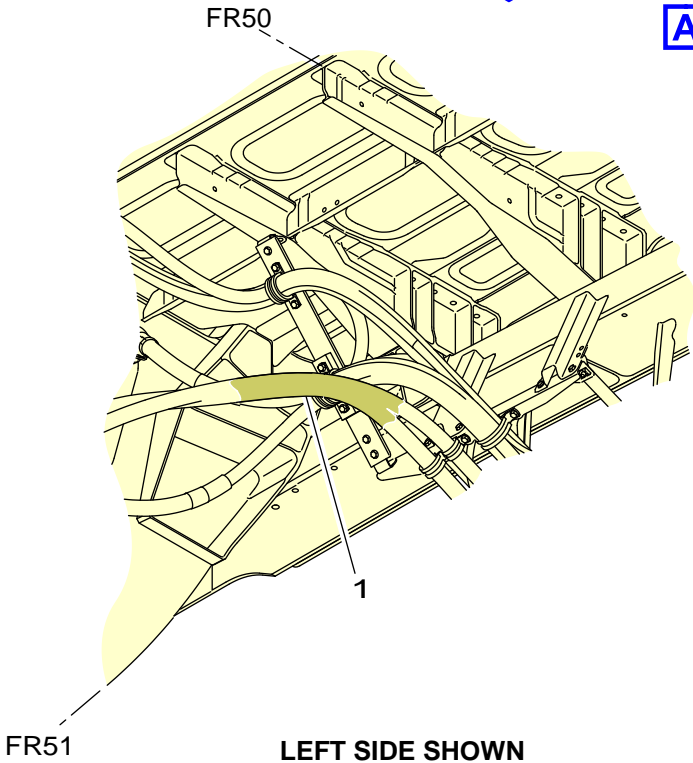
A220



LEFT SIDE SHOWN
RIGHT SIDE OPPOSITE

A

2



LEFT SIDE SHOWN
RIGHT SIDE OPPOSITE

B

LEGEND



Visual inspection

- 1. VFG power harness cable
- 2. Wheel bin

ICN-BD500-A-J534101-F-3AB48-51898-A-001-01

Figure 1 Inspection of the VFG Power-Feeder Harnesses and Wheel Bins

3 Job close-up

Note

The steps in the Job close-up section of this service bulletin, except for the return-to-service tests, are recommended steps. The steps give a recommendation to install components removed during the Job set-up. This recommendation is to make sure that the aircraft is safe and ready to return to service. Alternative steps can be used at the discretion of the operator.

3.1 Remove all tools, equipment, and unwanted materials from the work area.

4 Administrative

When PART A of this service bulletin is completed, record in the applicable aircraft records that you have completed PART A of this service bulletin.

Requirements after job completion

Required conditions

Table 6 Required conditions

Action/Condition	Data Module/Technical publication
None	

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PART B - Installation of Clamps and Routing Modification of the VFG Power-Feeder Harnesses - Accomplishment instruction

Applicability: 50010-50017, 55003-55023, 55025

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Table 1 References

Data Module/Technical Publication	Title
BD500-A-J12-00-00-01AAA-398A-A	Standard configuration for maintenance - Aircraft system configuration
BD500-A-J53-82-84-07AAA-520A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J53-82-84-08AAA-520A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J20-10-01-00AAA-720A-A	Wiring - Install procedure
BD500-A-J51-41-02-00AAA-078A-A	Bolts, screws, and related items - Fasteners
BD500-A-J20-31-00-00AAA-711A-A	Torquing of threaded fasteners - Tighten procedure
BD500-A-J53-82-84-07AAA-720A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Install procedure

Data Module/Technical Publication	Title
BD500-A-J53-82-84-08AAA-720A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Install procedure

Preliminary requirements

Required conditions

Table 2 Required conditions

Action/Condition	Data Module/Technical publication
None	

Support equipment

Table 3 Support equipment

Name	Identification/Reference	Quantity	Remark
None			

Consumables, materials, and expendables

Table 4 Consumables, materials, and expendables

Name	Identification/Reference	Quantity	Remark
None			

Spares

Table 5 Spares

Name	Identification/Reference	Quantity	Remark
None			

Safety conditions

None

Procedure

- 1 General information
Tolerance Information

LINEAR TOLERANCES		ANGULAR TOLERANCE
X.XX in. (X.X mm)	X.X in. (X mm)	± 0° 30"
± 0.03 in. (0.8 mm)	± 0.1 in. (3 mm)	

Note

The tolerances above apply to the dimensions given in this service bulletin, except when specified differently.

- 2 Job set-up

Note

The steps in the Job set-up section of this service bulletin are recommended steps. The steps give a recommendation to get access to the work area. This recommendation is to give a safe work area and to minimize possible damage to surrounding aircraft parts. Alternative steps can be used at the discretion of the operator.

- 2.1 Make sure that the aircraft is in standard configuration for maintenance (refer to AMP BD500-A-J12-00-00-01AAA-398A-A).

- 3 Procedure

- 3.1 If it is not possible to complete all the instructions in this service bulletin because of the configuration of the aircraft, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.
- 3.2 If necessary, remove the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-520A-A).
- 3.3 If necessary, remove the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-520A-A).
- 3.4 Modify the harness installation on the left side as follows:
Refer to Fig. 1
 - 3.4.1 Remove and discard the saddle clamp (10), screws (11) and washers (12) from the harnesses.
 - 3.4.2 Install the harnesses with a packer (1), and saddle clamp (4), screws (8), and washers (6) (refer to AMP BD500-A-J20-10-01-00AAA-720A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).
 - 3.4.3 Torque the screws (8) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).

CAUTION

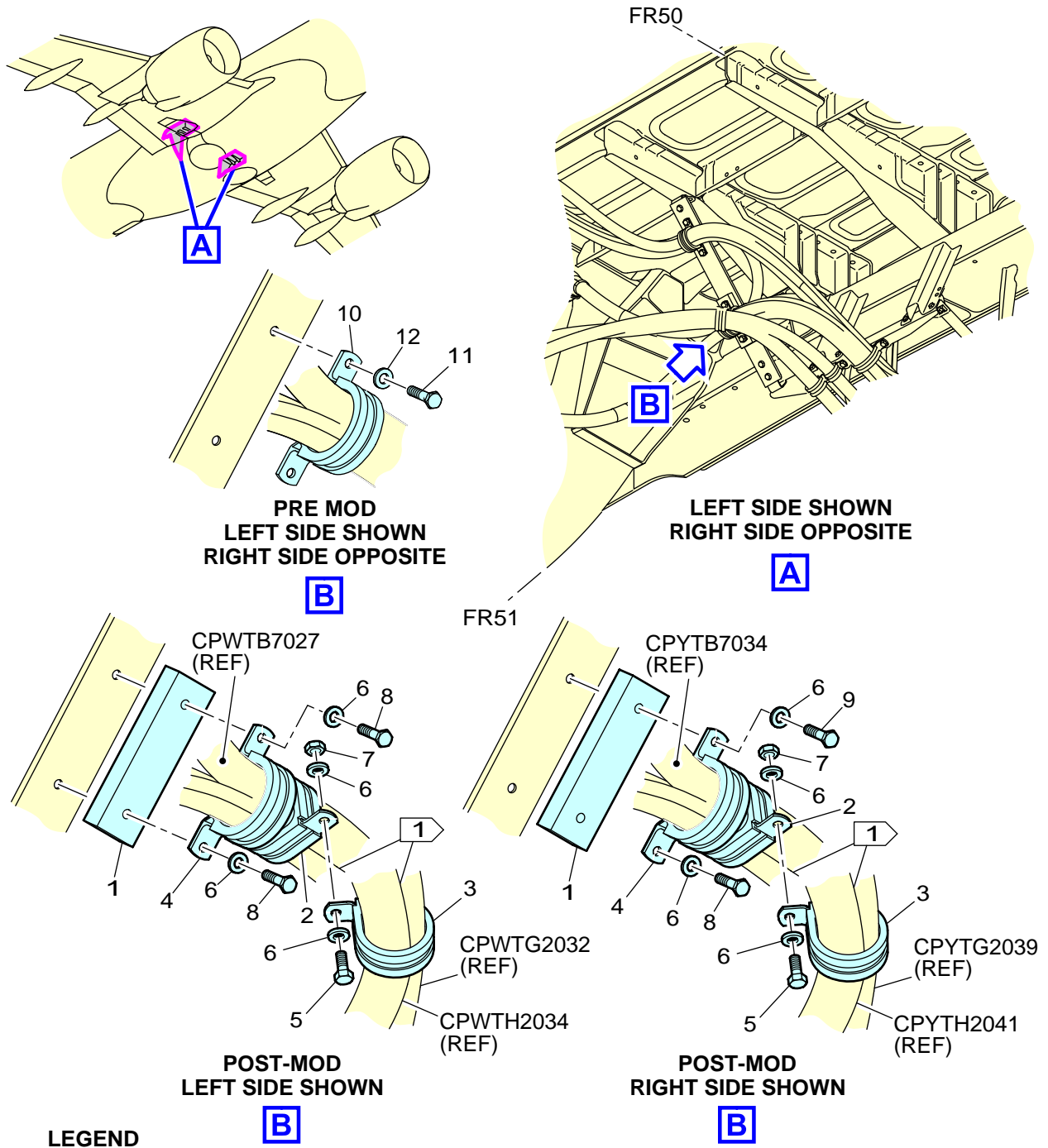
Be careful not to cause damage to the harness. The bend radius of the harness CPWTH2034 must not be less than 12 times its outside diameter.

- 3.4.4 Install a clamp (2) forward or aft of the clamp (4) around the harnesses and a clamp (3) around the Variable Frequency Generator (VFG) power-feeder harnesses CPWTG2032 and CPWTH2034. Attach the two clamps (2) and (3) with screw (5), washers (6), and one self-locking nut (7) (refer to AMP BD500-A-J20-10-01-00AAA-720A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).
- 3.4.5 Make sure that you keep a minimum clearance of 0.375 in. (9.53 mm) between the harnesses and the VFG power-feeder harnesses CPWTG2032 and CPWTH2034.
- 3.4.6 Torque screw (5) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
- 3.5 Modify the harness installation on the right side as follows:
Refer to Fig 1
 - 3.5.1 Remove and discard the saddle clamp (10), screws (11), and washers (12) from the harnesses.
 - 3.5.2 Install the harnesses with the packer (1), saddle clamp (4), screw (8), screw (9), and washers (6) (refer to AMP BD500-A-J20-10-01-00AAA-720A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).
 - 3.5.3 Torque screws (8) and (9) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).

CAUTION

Be careful not to cause damage to the harness. The bend radius of the harness CPYTH2041 must not be less than 12 times its outside diameter.

- 3.5.4 Install a clamp (2) forward or aft of the clamp (4) around the harnesses and a clamp (3) around the VFG power-feeder harnesses CPYTG2039 and CPYTH2041. Attach the two clamps (2) and (3) with screw (5), washers (6), and self-locking nut (7) (refer to AMP BD500-A-J20-10-01-00AAA-720A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).
- 3.5.5 Make sure that you keep a minimum clearance of 0.375 in. (9.53 mm) between the harnesses and the VFG power-feeder harnesses CPYTG2039 and CPYTH2041.
- 3.5.6 Torque the screw (5) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
- 3.6 Install the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-720A-A).
- 3.7 Install the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-720A-A).



LEGEND

- | | | |
|--|--------------------------------|------------------|
| 1. Packer C01384564-N0005 (CS100)
C01384564-N0006 (CS300) | 5. Screw B0201081CK3-9 | 10. Saddle clamp |
| 2. Clamp AS21919WCH22 | 6. Washer B0202033C0316K | 11. Screw |
| 3. Clamp AS21919WCH20 | 7. Self-locking nut NAS1291C3M | 12. Washer |
| 4. Saddle clamp TA09203CO22HC | 8. Screw B0201081CK3-16 | |
| | 9. Screw B0201081CK3-17 | |

NOTE

1 Make sure to keep a minimum clearance of 0.375 in. (9.53 mm) between both harnesses.

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Figure 1 Modification of the VFG Power Feeder-Harnesses Routing

4 Job close-up

Note

The steps in the Job close-up section of this service bulletin, except for the return-to-service tests, are recommended steps. The steps give a recommendation to install components removed during the Job set-up. This recommendation is to make sure that the aircraft is safe and ready to return to service. Alternative steps can be used at the discretion of the operator.

4.1 Remove all tools, equipment, and unwanted materials from the work area.

5 Administrative

When PART B of this service bulletin is completed, record in the applicable aircraft records that you have completed PART B of this service bulletin.

Requirements after job completion

Required conditions

Table 6 Required conditions

Action/Condition	Data Module/Technical publication
None	

PART C - Installation of Brackets and Supports and Routing Modification of the VFG Power- Feeder Harnesses - Accomplishment instruction

Applicability: 50010-50047, 55003-55070

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References

Table 1 References

Data Module/Technical Publication	Title
BD500-A-J12-00-00-01AAA-398A-A	Standard configuration for maintenance - Aircraft system configuration
BD500-A-J53-82-84-07AAA-520A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J53-82-84-08AAA-520A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Remove procedure
BD500-A-J51-42-06-00AAA-078A-A	Solid rivets - Fasteners

Data Module/Technical Publication	Title
BD500-A-J51-23-00-05AAA-259B-A	Wet installation of fasteners with sealing compound - Sealing procedures
BD500-A-J51-40-11-01AAA-652A-A	Metallic materials - Preparation of fastener holes
BD500-A-J51-23-00-09AAA-259B-A	Sealing mating surfaces - Sealing procedures
BD500-A-J51-25-16-00AAA-257A-A	Fluid-Resistant (FR) polyurethane topcoat - Paint and apply marking
BD500-A-J20-10-01-05AAA-913A-A	Installation of harness clamps - General maintenance procedure
BD500-A-J20-31-00-00AAA-711A-A	Torquing of threaded fasteners - Tighten procedure
BD500-A-J51-21-11-04AAA-259A-A	Corrosion inhibiting jointing compound - Corrosion prevention procedures
BD500-A-J51-41-02-00AAA-078A-A	Bolts, screws, and related items - Fasteners
BD500-A-J53-82-84-07AAA-720A-A	Wheel bin left, Wing To Body Fairing (WTBF) - Install procedure
BD500-A-J53-82-84-08AAA-720A-A	Wheel bin right, Wing To Body Fairing (WTBF) - Install procedure

Preliminary requirements

Required conditions

Table 2 Required conditions

Action/Condition	Data Module/Technical publication
None	

Support equipment

Table 3 Support equipment

Name	Identification/Reference	Quantity	Remark
None			

Consumables, materials, and expendables

Table 4 Consumables, materials, and expendables

Name	Identification/Reference	Quantity	Remark
None			

Spares

Table 5 Spares

Name	Identification/Reference	Quantity	Remark
None			

Safety conditions

None

Procedure

1 General information

Note

PART C of the service bulletin is classified as Mandatory or expected to be Mandatory by an Airworthiness Directive (AD). The Procedure section of the Accomplishment Instructions is Required for Compliance (RC) and must be done to comply with the AD. The Job set-up and Job close-up sections, with the exception of the return-to-service tests, are recommended and can be deviated from, done as part of other actions, or done with accepted methods different from those given in the service bulletin, as long as the RC section can be done and the aircraft can be put back into a serviceable condition.

Tolerance information

LINEAR TOLERANCES		ANGULAR TOLERANCE
X.XX in. (X.X mm)	X.X in. (X mm)	±0° 30'
±0.03 in. (0.8 mm)	±0.1 in. (3 mm)	

Note

The tolerances above apply to the dimensions given in this service bulletin, except when specified differently.

2 Job set-up

Note

The steps in the Job set-up section of this service bulletin are recommended steps. The steps give a recommendation to get access to the work area. This recommendation is to give a safe work area and to minimize possible damage to surrounding aircraft parts. Alternative steps can be used at the discretion of the operator.

- 2.1 Make sure that the aircraft is in standard configuration for maintenance (refer to AMP BD500-A-J12-00-00-01AAA-398A-A).
- 2.2 Remove the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-520A-A).
- 2.3 Remove the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-520A-A).

3 Procedure

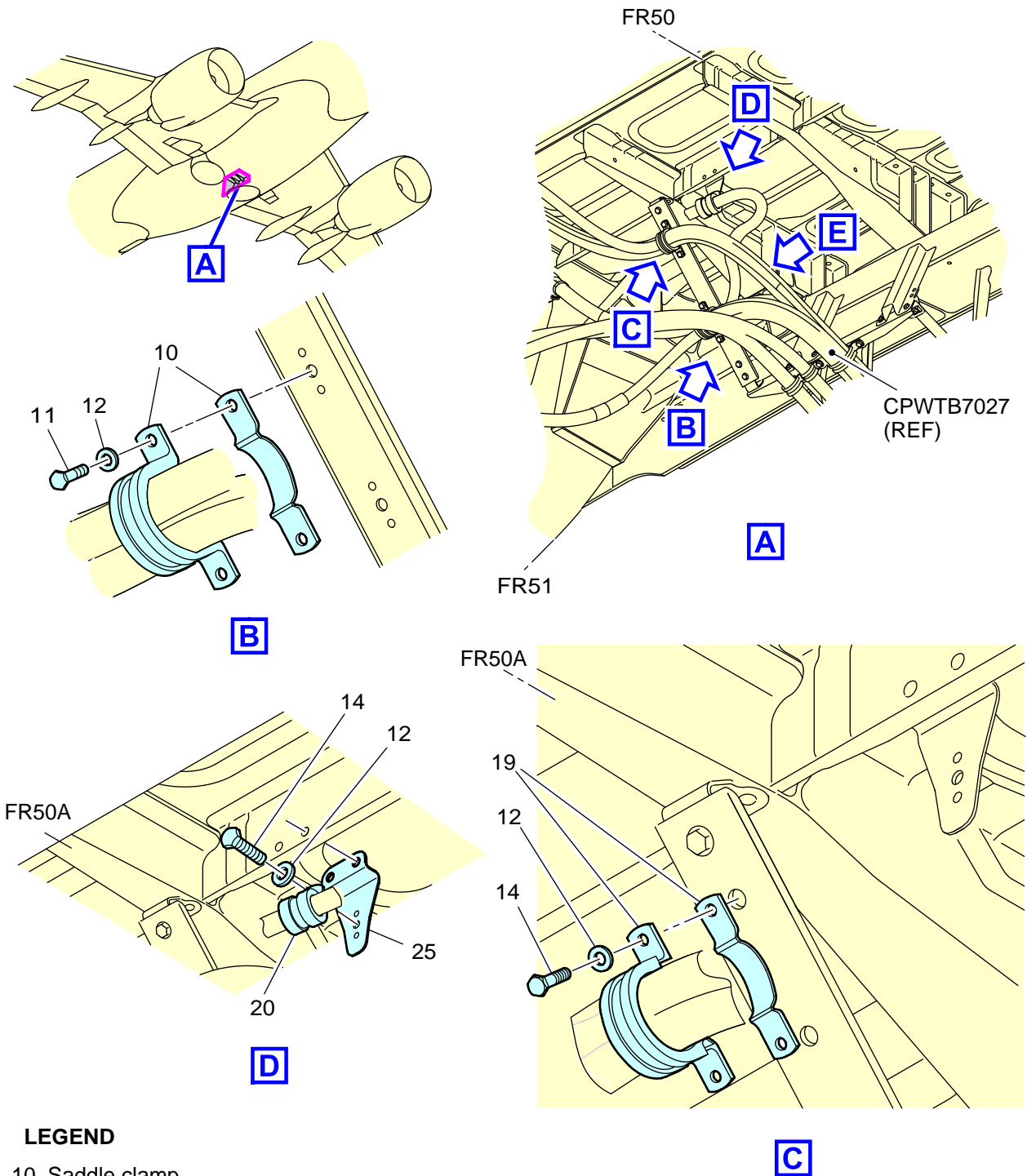
- 3.1 If it is not possible to complete all the instructions in this service bulletin because of the configuration of the aircraft, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.

Applicability: For aircraft 50010-50017, 55003-55023 and 55025 that did not do PART A and PART B of SB BD500-534101

- 3.2 Modify the harness installation on the left side as follows:

Refer to Fig. 1

- 3.2.1 Remove the saddle clamp (10), screws (11), and washers (12) that hold the harness CPWTB7027.
- 3.2.2 Discard the screws (11). Keep the saddle clamp (10) and the washers (12).
- 3.2.3 Remove and keep the saddle clamp (19), two screws (14), and two washers (12) that hold the harness CPWTB7027.
- 3.2.4 Remove the loop clamp (20), screw (14), and washer (12).
- 3.2.5 Discard the loop clamp (20). Keep the screw (14) and the washer (12).
- 3.2.6 Remove the electrical system support assembly (21), screws (14), and washers (22).
- 3.2.7 Discard the electrical system support assembly (21). Keep the screws (14) and the washers (22).
- 3.2.8 Remove and discard the two electrical system bracket assemblies (23) and (24) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).
- 3.2.9 Remove the electrical system bracket assembly (25) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).
- 3.2.10 Wet-install with sealant two rivets (17) in the redundant holes in the intercostal where the electrical system bracket assembly (25) was installed (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

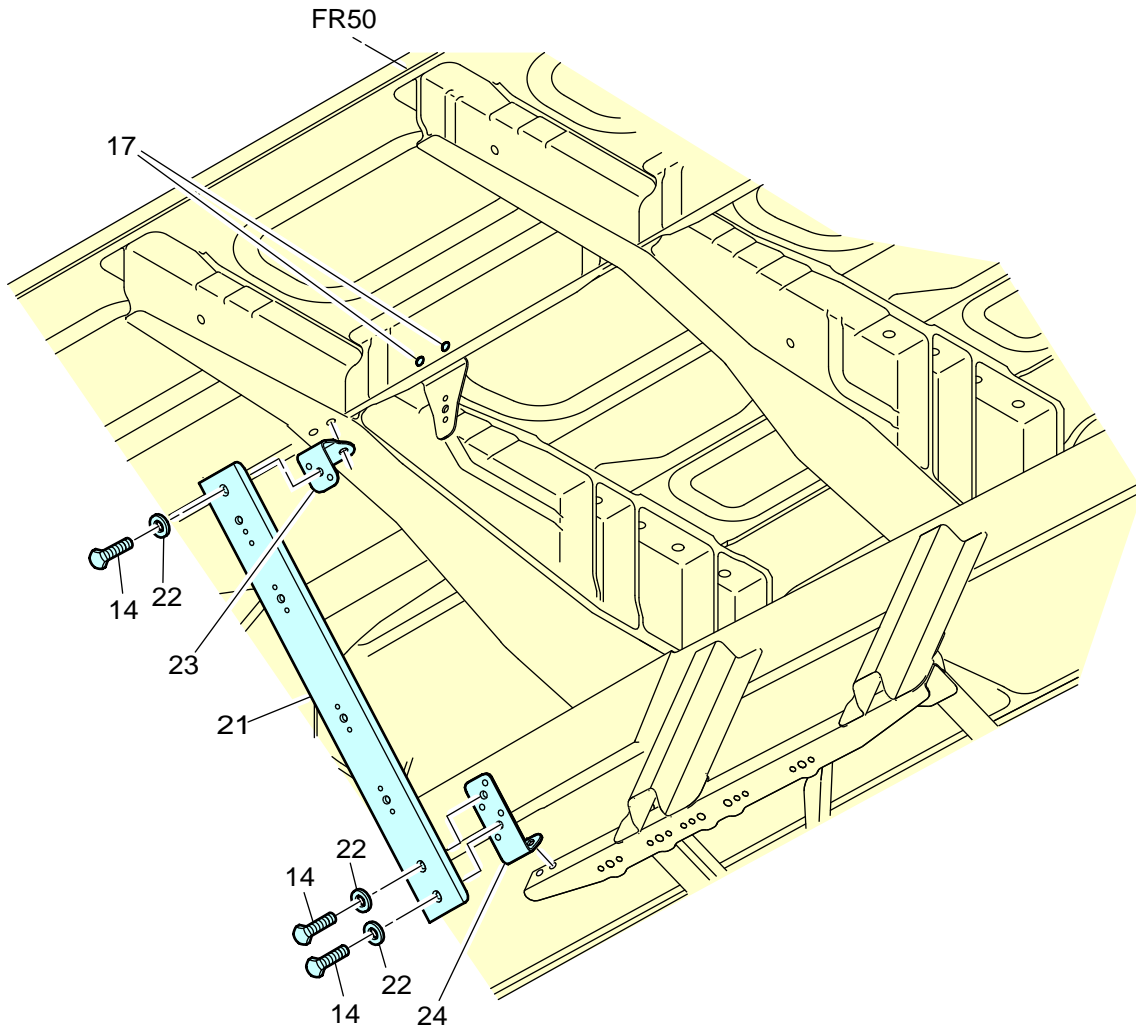


LEGEND

- 10. Saddle clamp
- 11. Screw, hex head
- 12. Washer, flat
- 14. Screw, hex head
- 19. Saddle clamp
- 20. Loop clamp
- 25. Bracket assembly

ICN-BD500-A-J534101-F-3AB48-57347-A-001-01

Figure 1 Left side - Routing modification of the power-feeder harnesses - Removal - (Sheet 1 of 2)



LEGEND

- 14. Screw, hex head
- 17. Rivet, solid MS20470AD4-4A
- 21. Support assembly
- 22. Washer, flat
- 23. Bracket assembly
- 24. Bracket assembly

ICN-BD500-A-J534101-F-3AB48-57348-A-002-01

Figure 1 Left side - Routing modification of the power-feeder harnesses - Removal - (Sheet 2 of 2)

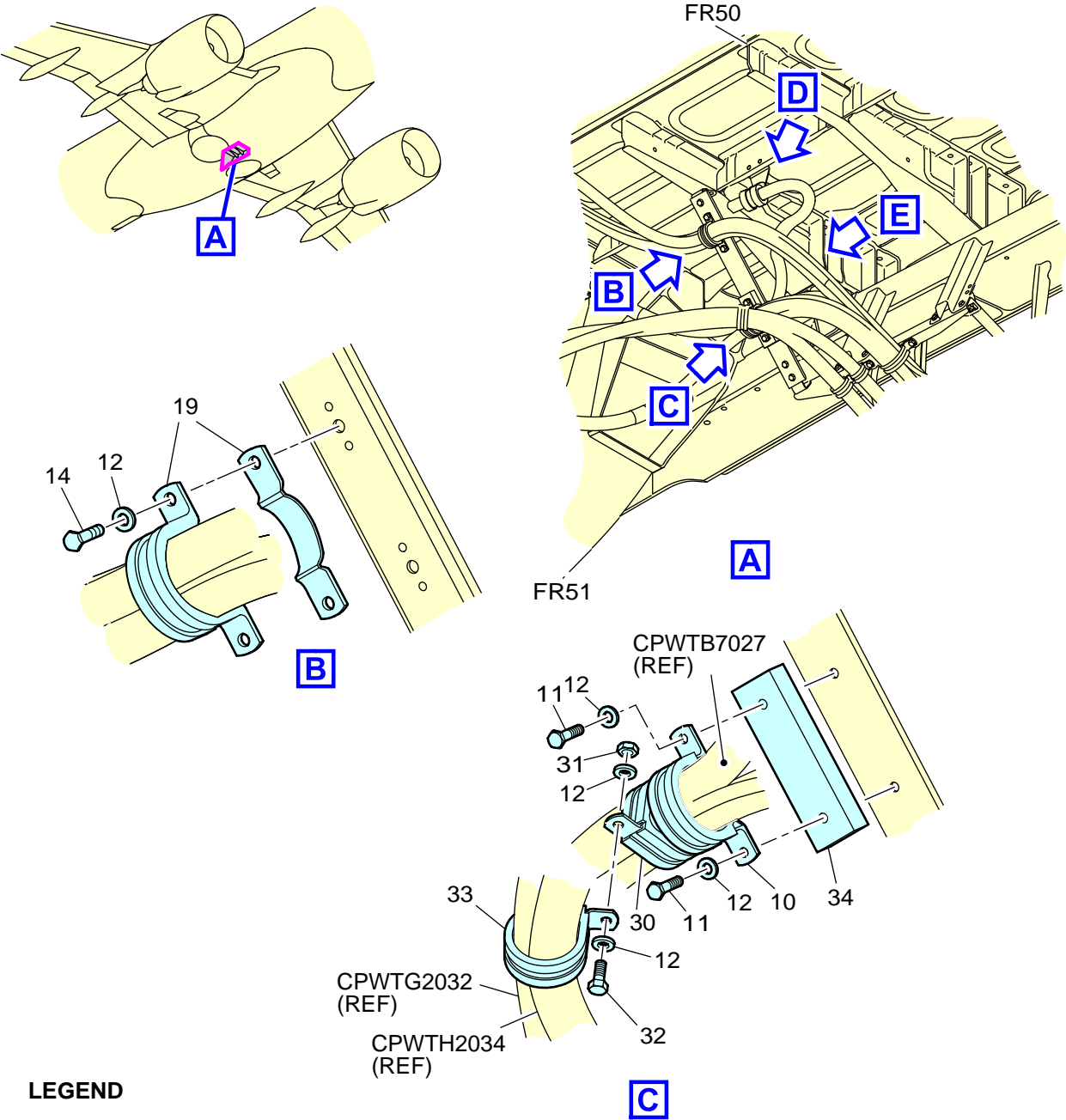
Applicability: For aircraft 50010-50017, 55003-55023, 55025 that did PART A and PART B of SB BD500-534101, and 50018 thru 50047, and 55024, 55026 thru 55070

3.3 Modify the harness installation on the left side as follows:

Refer to Fig. 2

- 3.3.1 Remove the clamp (30), clamp (33), nut (31), screw (32), and washers (12) that hold the harnesses CPWTH2034 and CPWTG2032.
- 3.3.2 Discard the clamp (30), clamp (33), nut (31), and screw (32). Keep the washers (12).
- 3.3.3 Remove the saddle clamp (10), screws (11), packer (34), and washers (12) that hold the harness CPWTB7027.
- 3.3.4 Discard the screws (11) and packer (34). Keep the clamp (10) and the washers (12).
- 3.3.5 Remove and keep the saddle clamp (19), two screws (14), and two washers (12) that hold the harness CPWTB7027.
- 3.3.6 Remove the clamp (20), screw (14), and washer (12).
- 3.3.7 Discard the clamp (20). Keep the screw (14) and the washer (12).
- 3.3.8 Remove the electrical system support (21), screws (14), and the washers (22).
- 3.3.9 Discard the electrical system support assembly (21). Keep the screws (14) and the washers (22).
- 3.3.10 Remove and discard the two electrical system bracket assemblies (23) and (24) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).
- 3.3.11 Remove the electrical system bracket assembly (25) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).
- 3.3.12 Wet-install with sealant two rivets (17) in the redundant holes of the electrical system bracket (25) on the intercostal (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

A220



LEGEND

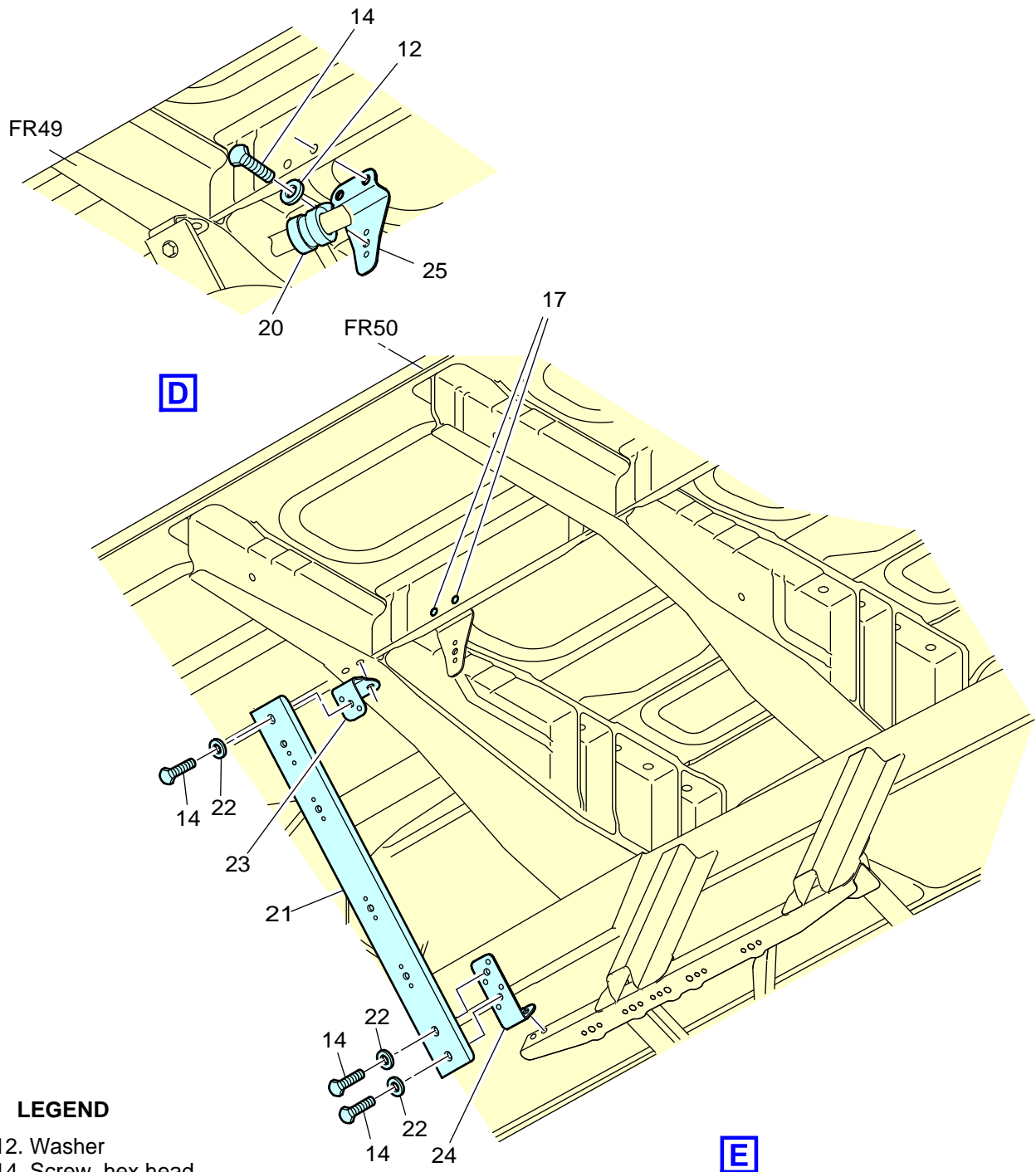
- 10. Saddle clamp
- 11. Screw
- 12. Washer, flat
- 14. Screw, hex head
- 19. Saddle clamp
- 30. Loop clamp
- 31. Self-locking nut
- 32. Screw
- 33. Loop clamp
- 34. Packer

ICN-BD500-A-J534101-F-3AB48-57351-A-001-01

Figure 2 Left side - Routing modification of the power-feeder harnesses - Removal - (Sheet 1 of 2)

See applicability on the first page of the DM
BD500-A-J53-41-01-03AAA-933A-A

BD500-A-J53-41-01-03AAA-933A-A
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LEGEND

- 12. Washer
- 14. Screw, hex head
- 17. Rivet, solid MS20470AD4-4A
- 20. Loop clamp
- 21. Support assembly
- 22. Washer, flat
- 23. Bracket assembly
- 24. Bracket assembly
- 25. Bracket assembly

ICN-BD500-A-J534101-F-3AB48-57352-A-002-01

Figure 2 Left side - Routing modification of the power-feeder harnesses - Removal - (Sheet 2 of 2)

Applicability: 50010-50047, 55003-55070

- 3.4 Install the electrical system brackets, supports and harnesses on the left side as follows:

Refer to Fig. 3

- 3.4.1 Install the electrical system bracket assembly (3) as follows:

3.4.1.1 Temporarily install the electrical system bracket assembly (3) using the existing holes where the electrical system bracket assembly (23) was installed on the frame 50A (refer to Fig. 1 or Fig. 2).

3.4.1.2 Backdrill two holes from FR50A through the electrical system bracket assembly (3) to a diameter of between 0.128 and 0.132 in. (3.25 to 3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).

3.4.1.3 Remove the electrical system bracket assembly (3).

3.4.1.4 Remove burrs and sharp edges on the electrical system bracket assembly (3) and FR50A.

3.4.1.5 Apply faying surface sealant to the electrical system bracket assembly (3) (refer to ASRP BD500-A-J51-23-00-09AAA-259B-A).

3.4.1.6 Wet-install with sealant the electrical system bracket assembly (3) on the FR50A with two rivets (13) wet with sealant (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

3.4.1.7 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (13) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).

- 3.4.2 Install the electrical system support assembly (7) as follows:

3.4.2.1 Temporarily install the electrical system support assembly (7) in the existing holes where the electrical system bracket assembly (24) was installed on the electrical system bracket installation at FR50A (refer to Fig. 1 or Fig. 2).

3.4.2.2 Backdrill two holes from the electrical system bracket installation through the electrical system support assembly (7) to a diameter of between 0.128 (3.25 mm) and 0.132 in. (3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).

3.4.2.3 Remove the electrical system support assembly (7).

3.4.2.4 Remove burrs and sharp edges on the electrical system support (7) and the electrical system bracket installation.

3.4.2.5 Wet-install with sealant the electrical system support assembly (7) on FR50A with two rivets (17) wet with sealant (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

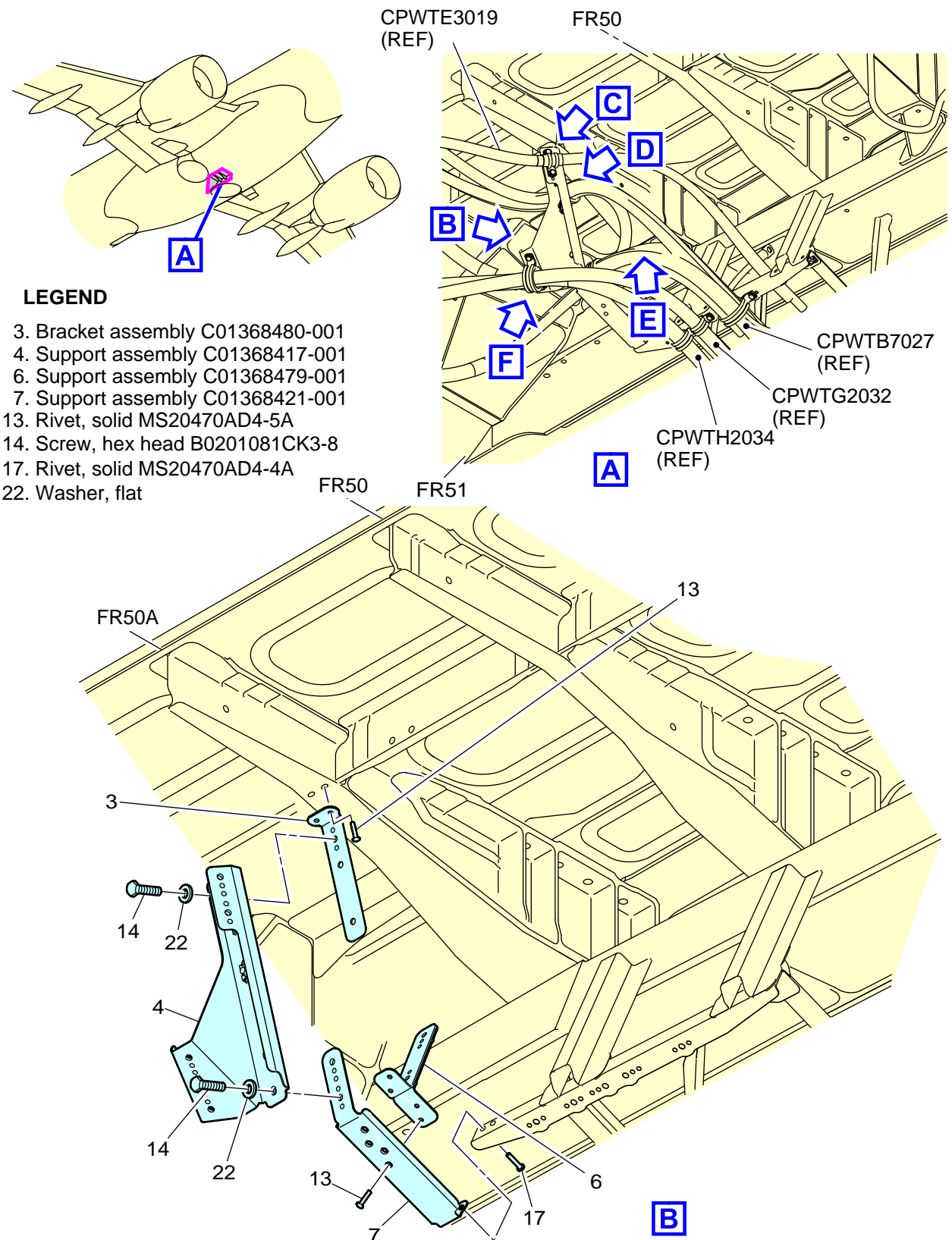
-
- 3.4.2.6 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (17) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).
 - 3.4.3 Install the electrical system support assembly (6) as follows:
 - 3.4.3.1 Temporarily install the electrical system support assembly (6) in the pilot holes on the support assembly (7).
 - 3.4.3.2 Open four holes in the electrical system support assembly (6) through the electrical system support assembly (7) to a diameter of between 0.128 (3.25 mm) and 0.132 in. (3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).
 - 3.4.3.3 Remove the electrical system support assembly (6).
 - 3.4.3.4 Remove burrs and sharp edges on the electrical system support assembly (6).
 - 3.4.3.5 Wet-install with sealant the electrical system support assembly (6) on the support assembly (7) with four rivets (13) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).
 - 3.4.3.6 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (13) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).
 - 3.4.4 Install the harness CPWTB7027 on the electrical system support assembly (6) as follows:
 - 3.4.4.1 Install the harness CPWTB7027 on the electrical system support assembly (6) with the saddle clamp (10), washers (12), and screws (18) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
 - 3.4.4.2 Torque the screws (18) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
 - 3.4.5 Install the harness CPWTB7027 on the electrical system support assembly (4) as follows:
 - 3.4.5.1 Install the harness CPWTB7027 on the electrical system support assembly (4) with the saddle clamp (19), washers (12), and screws (18) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
 - 3.4.5.2 Torque the screws (18) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
 - 3.4.6 Install the electrical system support assembly (4) as follows:
 - 3.4.6.1 Wet-install with corrosion-inhibiting compound the electrical system support assembly (4) on the electrical system bracket assembly (3) and the electrical system support assembly (7) with the three washers (22) and three screws (14) (refer to AMP BD500-A-J51-21-11-04AAA-259A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).

- 3.4.6.2 Torque the screws (14) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).

CAUTION

Be careful not to cause damage to the harness. The bend radius of the harness CPWTH2034 must not be less than 12 times its outside diameter.

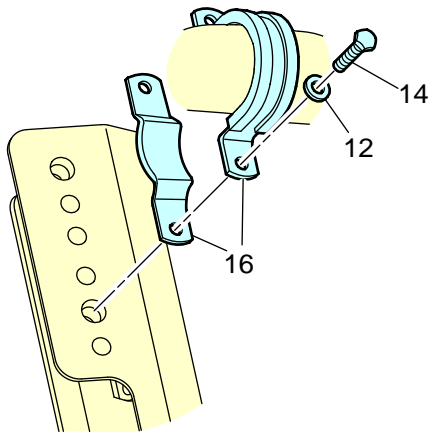
- 3.4.7 Install the harnesses CPWTG2032 and CPWTH2034 on the electrical system support assembly (4) as follows:
- 3.4.7.1 Install the harnesses CPWTG2032 and CPWTH2034 on the electrical system support assembly (4) with a clamp (15), screws (14), and washers (12) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
- 3.4.7.2 Torque the screw (14) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
- 3.4.8 Install the harness CPWTE3019 on the electrical system support assembly (4) as follows:
- 3.4.8.1 Install the harness CPWTE3019 on the electrical system support assembly (4) with a clamp (16), screws (14), and washers (12) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
- 3.4.8.2 Torque the screw (14) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
- 3.4.8.3 If the harness is still loose or could still rubbed against the wheel bin after the installation, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.



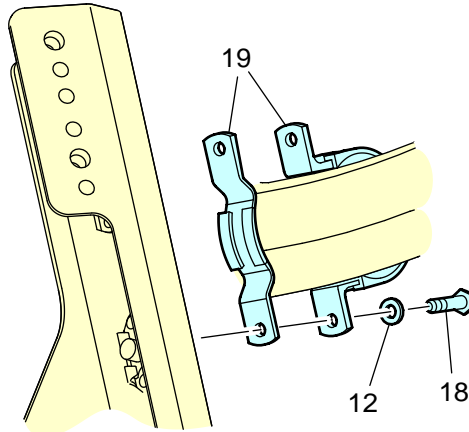
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Figure 3 Left side - Routing modification of the power-feeder harnesses - Installation - (Sheet 1 of 2)

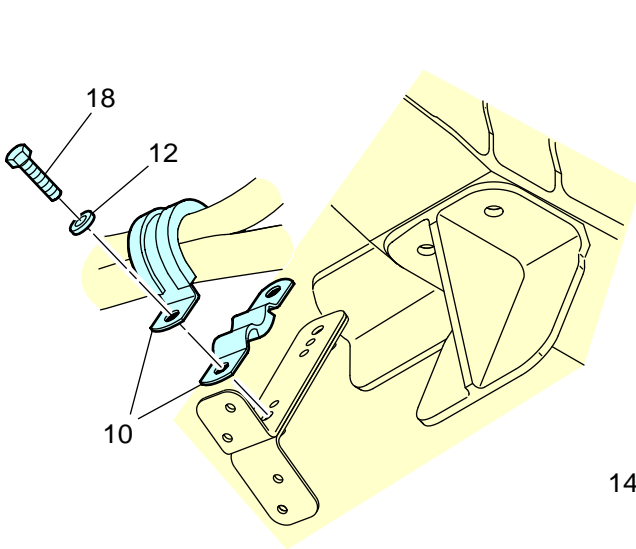
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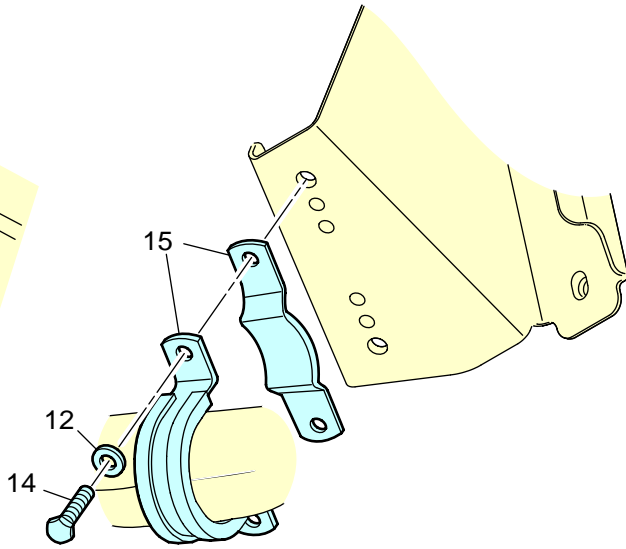
C



D



E



F

LEGEND

- 10. Saddle clamp
- 12. Washer, flat B0202033C0316K
- 14. Screw, hex head B0201081CK3-8
- 15. Saddle clamp TA09203CO20HC
- 16. Saddle clamp TA09201CO08HC
- 18. Screw, hex head B0201081CK3-9
- 19. Saddle clamp

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Figure 3 Left side - Routing modification of the power-feeder harnesses - Installation - (Sheet 2 of 2)

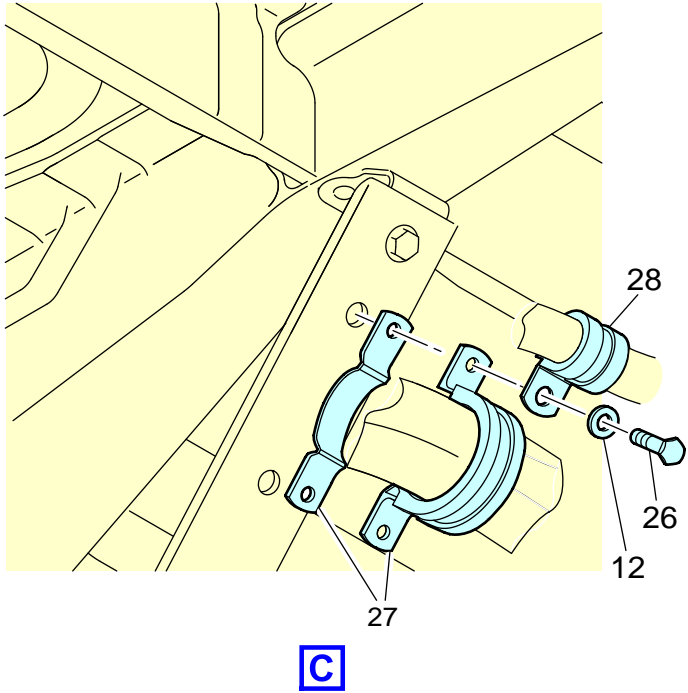
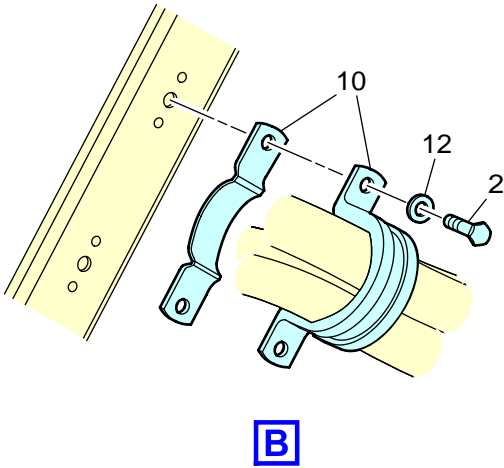
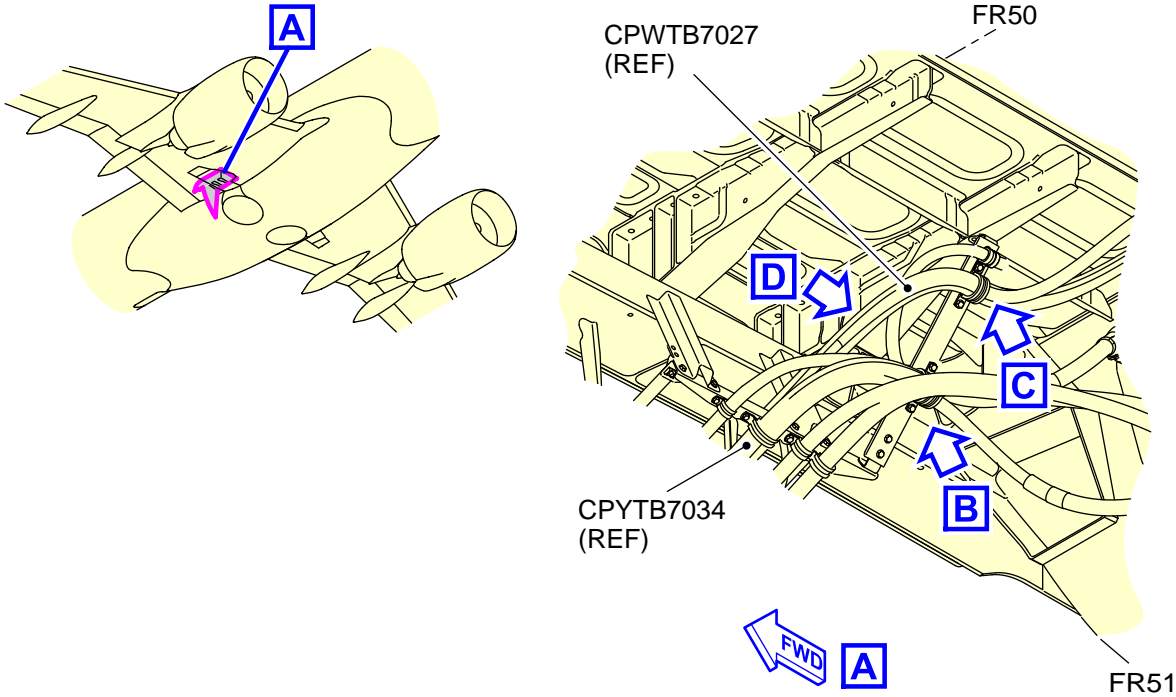
Applicability: For aircraft 50010-50019, 55003-55023, 55025 that did not do PART A and PART B of SB BD500-534101

3.5 Modify the harness installation on the right side as follows:

Refer to Fig. 4

- 3.5.1 Remove the saddle clamp (10), screws (2), and washers (12) that hold the harness CPYTB7034.
- 3.5.2 Discard the screws (2). Keep the clamp (10) and the washers (12).
- 3.5.3 Remove the saddle clamp (27), clamp (28), two screws (26), and two washers (12) that hold the harness CPWTB7027.
- 3.5.4 Discard the two screws (26). Keep the saddle clamp (27), clamp (28), and two washers (12).
- 3.5.5 Remove the electrical system support assembly (29), screws (14), and the washers (22).
- 3.5.6 Discard the electrical system support assembly (29). Keep the screws (14) and the washers (22).
- 3.5.7 Remove and discard the two electrical system brackets assemblies (23) and (24) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).

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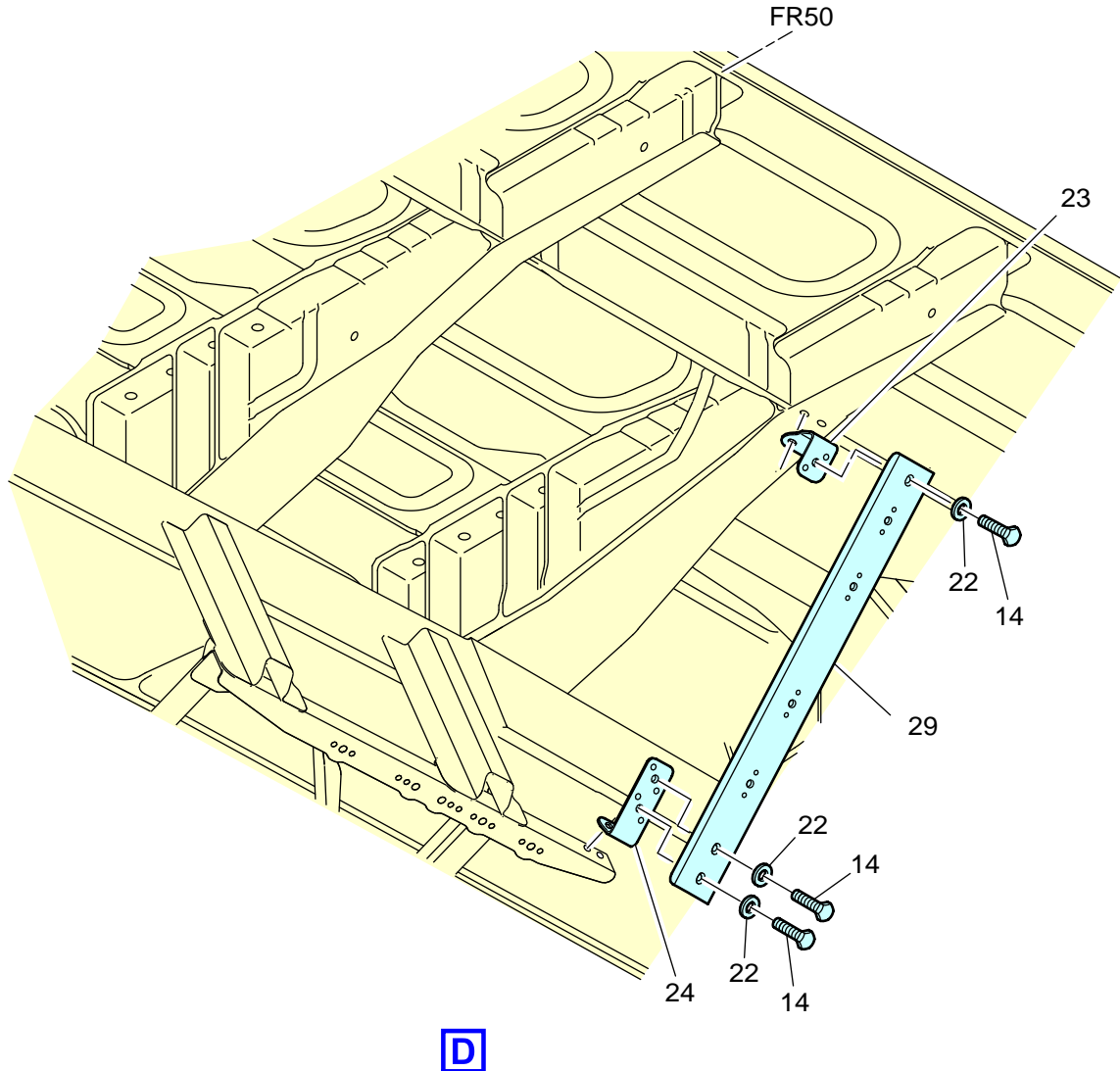


LEGEND

- 2. Screw
- 10. Saddle clamp
- 12. Washer, flat
- 26. Screw, hex head
- 27. Saddle clamp
- 28. Loop clamp

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Figure 4 Right side - Routing modification of the power-feeder harnesses - Removal - (Sheet 1 of 2)



LEGEND

- 14. Screw, hex head
- 22. Washer, flat
- 23. Bracket assembly
- 24. Bracket assembly
- 29. Support assembly

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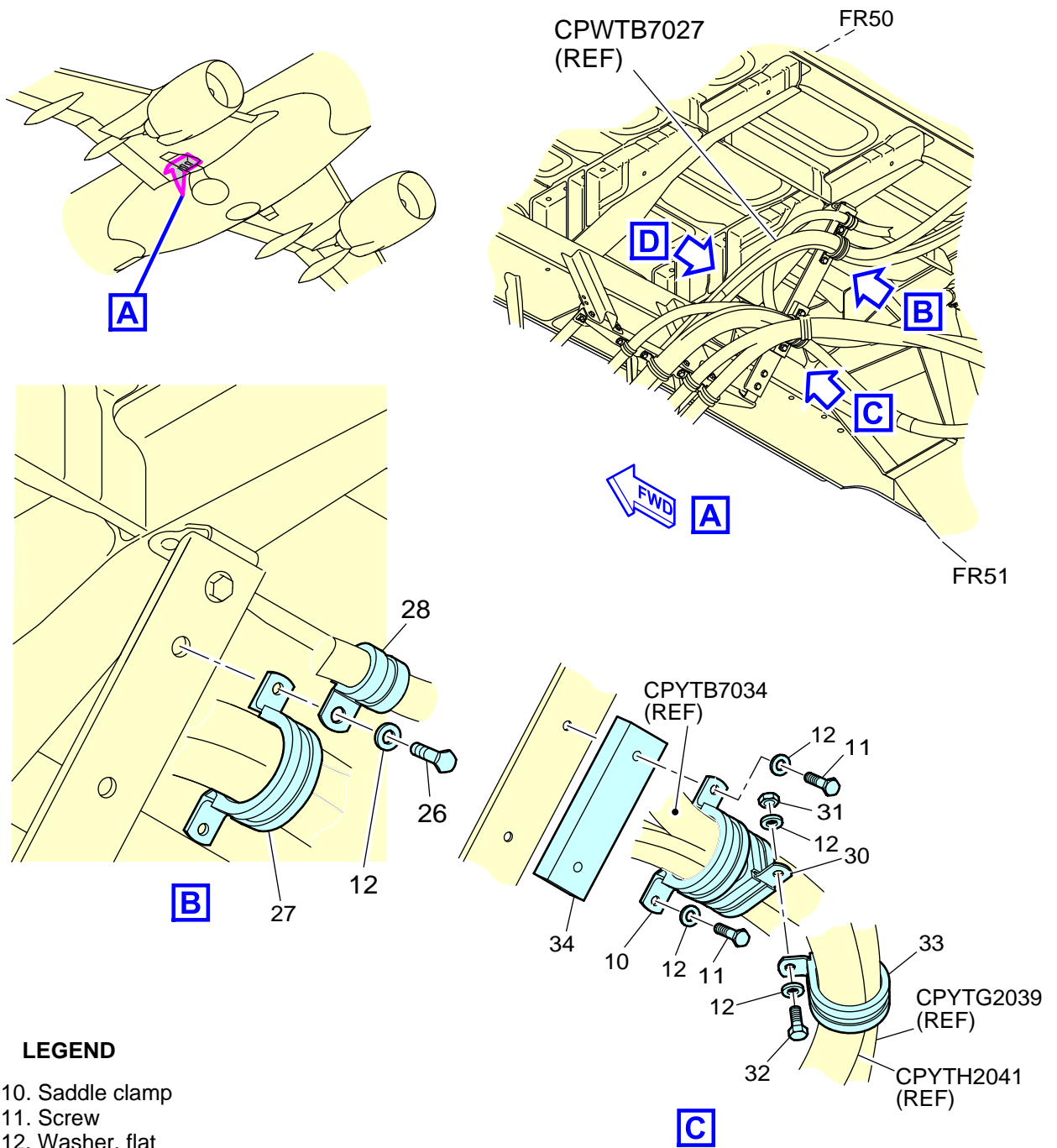
Figure 4 Right side - Routing modification of the power-feeder harnesses - Removal - (Sheet 2 of 2)

Applicability: For aircraft 50010-50017, 55003-55023, 55025 that did PART A and PART B of SB BD500-534101, and 50018 thru 50047, and 55024, 55026 thru 55070

3.6 Modify the harness installation on the right side as follows:

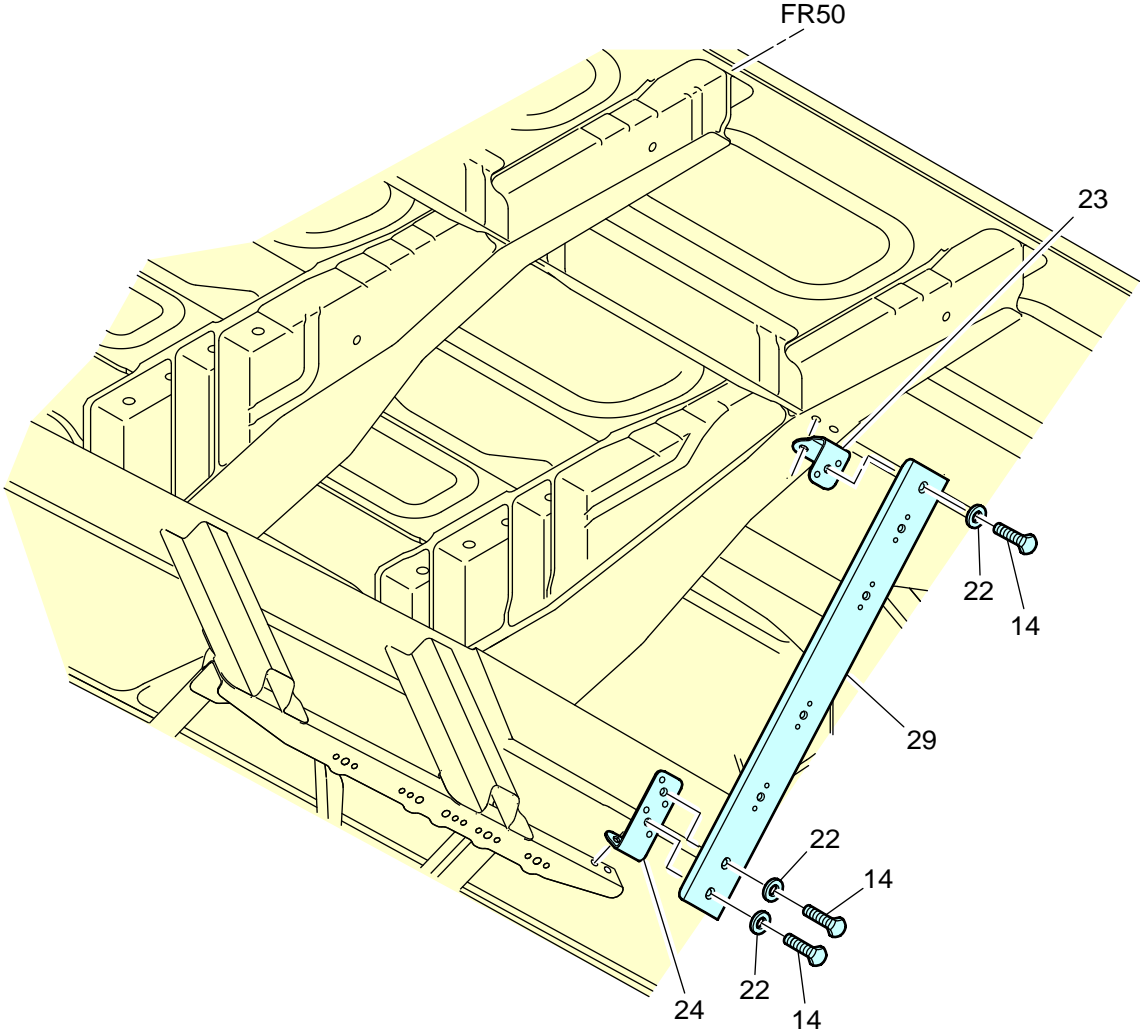
Refer to Fig. 5

- 3.6.1 Remove the clamp (30), clamp (33), nut (31), screw (32), and washers (12) that hold the harnesses CPYTG2039 and CPYTH2041.
- 3.6.2 Discard the clamp (30), clamp (33), nut (31), and screw (32). Keep the washers (12).
- 3.6.3 Remove the saddle clamp (10), screws (11), packer (34), and washers (12) that hold the harness CPYTB7034.
- 3.6.4 Discard the screws (11) and the packer (34). Keep the saddle clamp (10) and washers (12).
- 3.6.5 Remove the saddle clamp (27), saddle clamp (28), two screws (26), and two washers (12) that hold the harness CPWTB7027.
- 3.6.6 Discard the two screws (26). Keep the saddle clamp (27), saddle clamp (28), and two washers (12)
- 3.6.7 Remove the electrical system support assembly (29), screws (14), and washers (22).
- 3.6.8 Discard the electrical system support assembly (29). Keep the screws (14) and the washers (22).
- 3.6.9 Remove and discard the two electrical system brackets assemblies (23) and (24) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A).



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Figure 5 Right side - Routing modification of the power-feeder harnesses - Removal - (Sheet 1 of 2)



LEGEND

- 14. Screw, hex head
- 22. Washer, flat
- 23. Bracket assembly
- 24. Bracket assembly
- 29. Support assembly

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Figure 5 Right side - Routing modification of the power-feeder harnesses - Removal - (Sheet 2 of 2)

Applicability: 50010-50047, 55003-55070

3.7 Install the electrical system brackets, supports and harnesses on the right side as follows:

Refer to Fig. 6

3.7.1 Install the electrical system bracket assembly (5) as follows:

3.7.1.1 Temporarily install the electrical system bracket assembly (5) in the existing holes where the electrical system bracket assembly (23) was installed on the FR50A (refer to Fig. 4 or Fig. 5).

3.7.1.2 Backdrill two holes from FR50A through the electrical system bracket assembly (5) to a diameter of between 0.128 (3.25 mm) and 0.132 in. (3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).

3.7.1.3 Remove the electrical system bracket assembly (5).

3.7.1.4 Remove burrs and sharp edges on the electrical system bracket assembly (5) and FR50A.

3.7.1.5 Apply faying surface sealant to the electrical system bracket assembly (5) (refer to ASRP BD500-A-J51-23-00-09AAA-259B-A).

3.7.1.6 Wet-install with sealant the electrical system bracket assembly (5) on FR50A with two rivets (13) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

3.7.1.7 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (13) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).

3.7.2 Install the electrical system support assembly (8) as follows:

3.7.2.1 Temporarily install the electrical system bracket assembly (8) in the existing holes where the electrical system bracket assembly (24) was installed on the FR50A (refer to Fig. 4 or Fig. 5).

3.7.2.2 Backdrill two holes from the electrical system bracket installation through the electrical system support assembly (8) to a diameter of between 0.128 (3.25 mm) and 0.132 in. (3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).

3.7.2.3 Remove the electrical system support assembly (8).

3.7.2.4 Remove burrs and sharp edges on the electrical system support assembly (8) and the electrical system bracket installation.

3.7.2.5 Wet-install with sealant the electrical system support assembly (8) on FR50A with two rivets (17) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).

3.7.2.6 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (17) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).

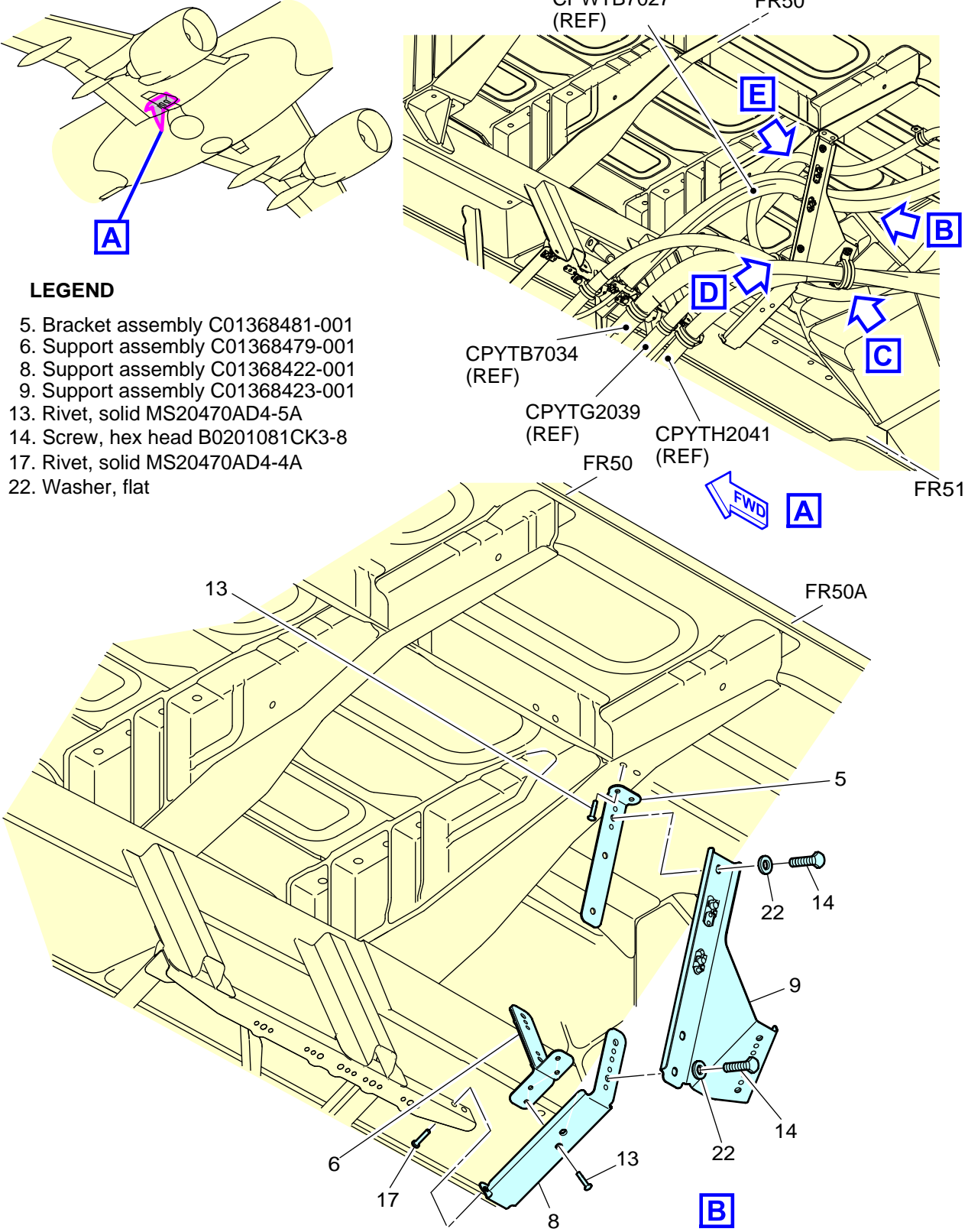
-
- 3.7.3 Install the electrical system support assembly (6) as follows:
 - 3.7.3.1 Temporarily install the electrical system support assembly (6) in the using the pilot holes on the support assembly (8).
 - 3.7.3.2 Open four holes in the electrical system support assembly (6) to a diameter of between 0.128 (3.25 mm) and 0.132 in. (3.35 mm) (refer to ASRP BD500-A-J51-40-11-01AAA-652A-A).
 - 3.7.3.3 Remove the electrical system support assembly (6).
 - 3.7.3.4 Remove burrs and sharp edges on the electrical system support assembly (6).
 - 3.7.3.5 Wet-install with sealant the electrical system support assembly (6) on the support assembly (8) with four rivets (13) (refer to ASRP BD500-A-J51-42-06-00AAA-078A-A and ASRP BD500-A-J51-23-00-05AAA-259B-A).
 - 3.7.3.6 Apply a layer of fluid-resistant polyurethane topcoat to the rivets (13) to agree with the color of the adjacent area (refer to ASRP BD500-A-J51-25-16-00AAA-257A-A).
 - 3.7.4 Install the harness CPYTB7034 on the electrical system support assembly (6) as follows:
 - 3.7.4.1 Install the harness CPYTB7034 on the electrical system support assembly (6) with the saddle clamp (10), washers (12), and screws (18) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
 - 3.7.4.2 Torque the screws (18) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
 - 3.7.5 Install the harnesses CPWTB7027 and CPYTB7034 on the electrical system support assembly (9) as follows:
 - 3.7.5.1 Install the harnesses CPWTB7027 and CPYTB7034 on the electrical system support assembly (9) with the saddle clamp (27), saddle clamp (28), washers (12), and screws (35) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
 - 3.7.5.2 Torque the screws (35) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
 - 3.7.6 Install the electrical system support assembly (9) as follows:
 - 3.7.6.1 Wet-install with corrosion-inhibiting compound the electrical system support assembly (9) on the electrical system bracket assembly (5) and the electrical system support assembly (8) with the three washers (22) and three screws (14) (refer to AMP BD500-A-J51-21-11-04AAA-259A-A and ASRP BD500-A-J51-41-02-00AAA-078A-A).
 - 3.7.6.2 Torque the screws (14) from 25 to 35 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).

CAUTION

Be careful not to cause damage to the harness. The bend radius of the harness CPYTH2041 must not be less than 12 times its outside diameter.

- 3.7.7 Install the harnesses CPYTG2039 and CPYTH2041 on the electrical system support assembly (9) as follows:
- 3.7.7.1 Install the harnesses CPYTG2039 and CPYTH2041 on the electrical system support assembly (9) with a saddle clamp (15), screws (14), and washers (12) (refer to ESPP BD500-A-J20-10-01-05AAA-913A-A).
 - 3.7.7.2 Torque the screws (14) to 25.0 to 35.0 lbf•in (2.83 to 3.95 Nm) (refer to AMP BD500-A-J20-31-00-00AAA-711A-A).
 - 3.7.7.3 If the harness is still loose or could still rubbed against the wheel bin after the installation, speak with a CRC representative at 1-866-A220-CRC (1-866-222-0272) or 1-450-476-7676 or e-mail at A220_crc@abc.airbus for analysis and to get an approved disposition to complete this service bulletin.

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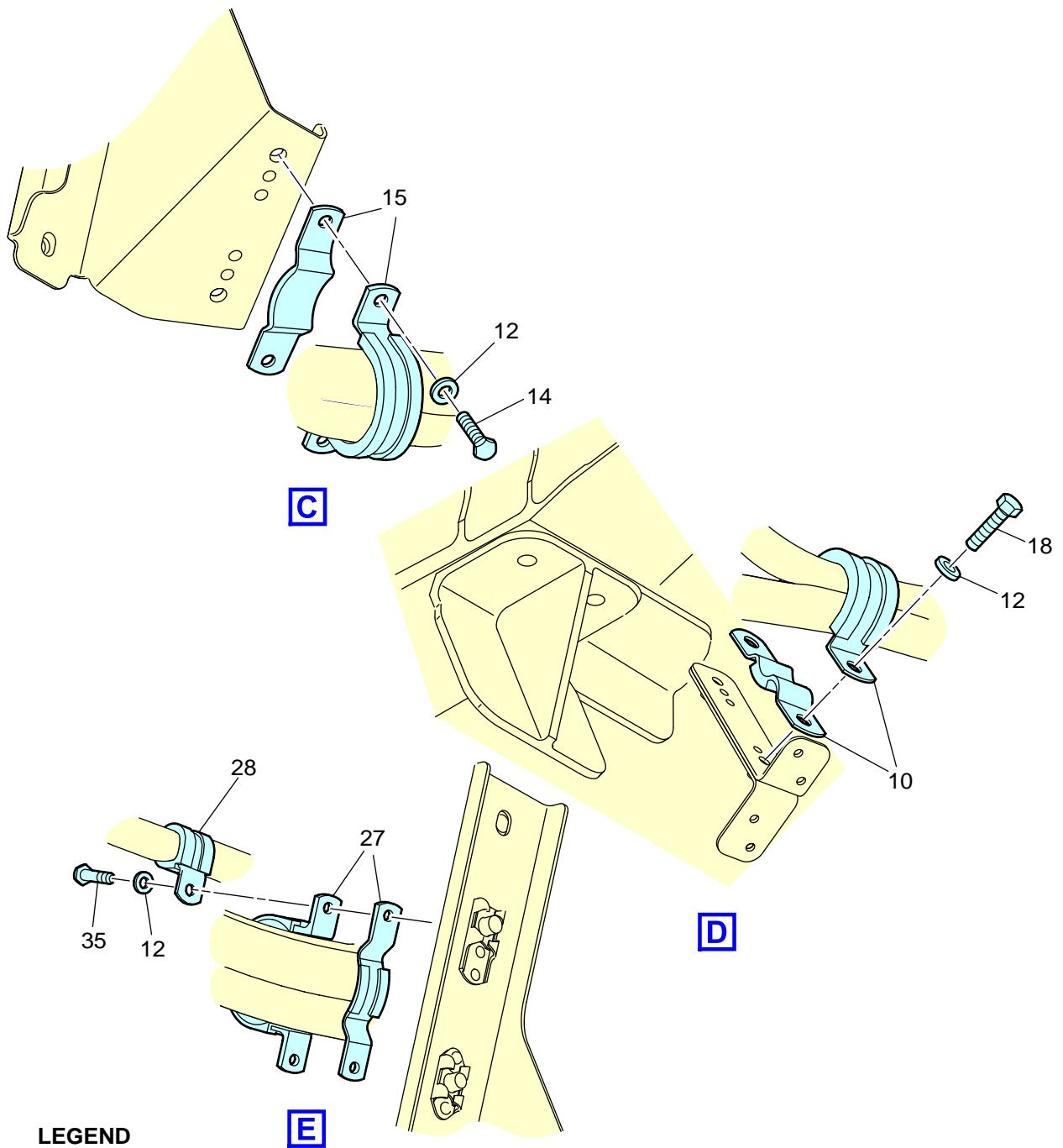


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Figure 6 Right side - Routing modification of the power-feeder harnesses - Installation - (Sheet 1 of 2)

See applicability on the first page of the DM
BD500-A-J53-41-01-03AAA-933A-A

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LEGEND

- 10. Saddle clamp
- 12. Washer, flat B0202033C0316K
- 14. Screw, hex head B0201081CK3-8
- 15. Saddle clamp TA09203CO20HC
- 18. Screw, hex head B0201081CK3-9
- 27. Saddle clamp
- 28. Saddle clamp
- 35. Screw, hex head B0201081CK3-10

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Figure 6 Right side - Routing modification of the power-feeder harnesses - Installation - (Sheet 2 of 2)

4 Job close-up

Note

The steps in the Job close-up section of this service bulletin, except for the return-to-service tests, are recommended steps. The steps give a recommendation to install components removed during the Job set-up. This recommendation is to make sure that the aircraft is safe and ready to return to service. Alternative steps can be used at the discretion of the operator.

- 4.1 Remove all tools, equipment, and unwanted materials from the work area.
- 4.2 Install the left wheel bin (refer to AMP BD500-A-J53-82-84-07AAA-720A-A).
- 4.3 Install the right wheel bin (refer to AMP BD500-A-J53-82-84-08AAA-720A-A).

5 Administrative

When PART C of this service bulletin is completed, record in the applicable aircraft records that you have completed PART C of this service bulletin.

When PART C of this service bulletin is completed, record the weight and balance data shown in the Weight and balance section of this service bulletin in the applicable document.

Requirements after job completion

Required conditions

Table 6 Required conditions

Action/Condition	Data Module/Technical publication
None	